

THIS IS RECONNAISSANCE MISSION NUMBER 2

HELIOCENTRIC EQUATORIAL COORDINATE SYSTEM EPOCH 1950.0

LAUNCH PLANET = EARTH

VENUS

ARRIVAL PLANET = JUPITER

INJECTION ALTITUDE = 200.00 KM

THIS IS RECONNAISSANCE MISSION NUMBER 3
HELIOCENTRIC EQUATORIAL COORDINATE SYSTEM EPOCH 1

LAUNCH PLANET = EARTH

VENUS

ARRIVAL PLANET = JUPITER

INJECTION ALTITUDE = 200.00 KM

SAVE CALLED AT THIS POINT***
LINES OUTPUT THIS JOB.

END OF OUTPUT FOR CFC9B*****

THIS IS RECONNAISSANCE MISSION NUMBER 4

HELIOCENTRIC EQUATORIAL COORDINATE SYSTEM EPOCH 1

LAUNCH PLANET = EARTH

VENUS

ARRIVAL PLANET = JUPITER

INJECTION ALTITUDE = 200.00 KM

SAVE CALLED AT THIS POINT***
LINES OUTPUT THIS JOB.

END OF OUTPUT FOR CF09B*****

THIS IS RECONNAISSANCE MISSION NUMBER 5

HELIOCENTRIC EQUATORIAL COORDINATE SYSTEM EPOCH

LAUNCH PLANET = EARTH

VENUS

ARRIVAL PLANET = JUPITER

INJECTION ALTITUDE = 200.00 KM

SAVE CALLED AT THIS POINT***
LINES CUTPUT THIS JCB.

END OF CUTPUT FOR CF09B*****

JOB NUMBER CF09B CONTINUATION OF SAVED JOB**

INTERPRETATION OF SYMBOLS

IV = INJECTION VELOCITY

RSI = RADIUS OF GRAVITATIONAL SPHERE OF INFLUENCE

HEV = HYPERBOLIC EXCESS VELOCITY

PPV = PLANETS POSITION VECTOR

VVVS,P = VEHICLES VELOCITY VECTOR W.R.T. SUN (AU/DAY) AND W.R.T. PLANET (KM/SEC)

VPVS,P = VEHICLES POSITION VECTOR W.R.T. SUN (AU) AND W.R.T. PLANET (KM)

THETA = TURN ANGLE OF VEHICLE ABOUT SUN ON ELLIPTICAL TRANSFER TRAJECTORY

PHI = ANGLE BETWEEN NORMAL TO TRANSFER PLANE AND NORMAL TO ECLIPTIC PLANE

A = SEMI-MAJOR AXIS

E = ECCENTRICITY

H-VECTORS GIVEN IN UNITS OF AU**2/DAY

DOCA = DISTANCE OF CLOSEST APPROACH TO PLANETS SURFACE

VOCA = VELOCITY AT CLOSEST APPROACH

TISI = TIME IN SPHERE OF INFLUENCE

DA = TOTAL DEFLECTION ANGLE OF VEHICLES VELOCITY VECTOR DUE TO PLANETS GRAVITATIO

TFT = TOTAL FLIGHT TIME
LINES OUTPUT THIS JOB.

END OF OUTPUT FOR CF098*****